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AVIATION

The Older American Aeronautical Magazine

EDWARD F. WATSON, Editor

LEON F. STOL, Publishing Director

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AVIATION

FOR JUNE 1933

The equipment of air forces

FRANCE PART TWO. BOMBERS, MULTIPLE FIGHTERS, AND NAVAL AIRCRAFT

By John Jay Lee

IN FRANCE, as in other European countries, bombing aircraft are subdivided into several groups. The heavy bomber has attracted a great deal of attention, but it has not by any means forced the lighter types from the field. As in most European countries, the light and heavy classes deliver loads each other in performance as well as in size and capacity, and are somewhat distinguished by the type of operation in which it is presumed that they would typically be employed—as day and night bombers, respectively.

[The distinction between day and night bombardment has played relatively little part in the selection and specification of material for the American forces in the last six or eight years. In Europe, however, perhaps because of the anticipation that bombing operations will have to cross over battle-lines successfully defended by anti-aircraft artillery, impossible to traverse by day except at a very high altitude, the distinction remains very important—Ed.]

Day bombers

Day bombing is closely allied to observation in the development of its equipment. At the end of the War, the Breguet 1932 was used almost exclusively. A slightly enlarged version of the Delagrange 1922 observation type, and having the same engine (a 500 hp Renault), it carried 573 lb. of bombs, fuel, and crew.

The armament consisted of one fixed gun for the pilot, and twin guns on a swing for the observer. It took 47 minutes to climb to 16,000 ft., and the speed at that height was 90 mph.

In 1923 the Breguet 19 (observation plane) was equipped for day bombing, and later provided with an engine of

some 450 hp. Later, in 1925, the most structure was strengthened to meet the new load factors which entered into force in that year. The Breguet 1932 has a normal bomb load of 500 lb. with a range of 500 mi. (Its speed at sea level is 100 mph, falling off to 124 mph at 16,000 ft., while the time of climb to 16,000 ft. is 25 minutes. These figures are for a total weight of 5,000 lb. at which 2,500 lb. is useful load.)

Concurrently there was developed the Potez 251E, likewise derived from the observation type. Although the Potez was almost entirely of wood construction, in some respects it was superior to the Breguet, as with the latter bomb-sighting was made somewhat difficult by the large forward section fuselage.

In 1924 a more ambitious bomber made its appearance and it still very much tops the same list. It was the S.1.C.M.—Armstrong designed biplane equipped with the 650 hp. engine-cylinder Lorraine engine, finally put into production in 1927 under the designation "Type 122." A three-seater carrying 1,500 lb. of bombs, it has a range of 655 miles. The bomber can ascend vertically to its maximum, the third member of the crew being a machine gunner for tail protection. Reconnaissance armament has been obtained with this airplane, the bomber carrying through a long run in the fuselage nose. The S.1.C.M.—Armstrong 122 has a complete metal structure, aluminum. Its characteristics are wing area, 593 sq ft., weight equipped, 5,800 lb., useful load, 2,001 lb., total weight, 6,800 lb., speed at 15,000 ft., 122 mph.; climb to 15,000 ft., 27 minutes.

In 1922 examples of this airplane were built with the Lorraine "Ocean" and the engine-cylinder Hispano-Suiza engines, developing 550 and 1,125 hp. respectively. It is probable that the machine eventually developed again as the most desirable multiple fighter (a category described below) will also be chosen to perform day bombing duties.

"Who walks by night"

The night bombing airplane in service in 1933 comprised the twin-engine Potez 25 and Caudron 25 and the Breguet 16 with a single 300-hp Renault. The last-named carried a load of 1,212 lb. of bombs, and it managed to reach a speed of 76 mph at 15,000 ft.

In 1922 there appeared the Loire & Olivier L.O.7 biplane with two 300-hp. Hispano-Suiza engines, and an 18-hp. Parnassus biplane with four 275-hp. engines and a crew of four men.

In 1923 the great achievement, first Schneider, at Le Croisic, brought out to the design of Captain Lefevre (who was responsible for the famous Potez-Lefevre biplane) an almost 11-ton biplane bomber with four engines.

In 1924, the firm of Loire & Olivier produced the L.O.12 biplane with two 300-hp. Lorraine engines, but this airplane was at the time constructed by the Parnassus "Goliath" having two 250-hp. engines. The Goliath was transformed into the P.O.1, to carry a ton of bombs, by Strick 425-hp. General-Rhône engines, and that version remained in service until 1931. After the Parnassus 65 was finally withdrawn, night bombing became the exclusive prerogative of the Loire & Olivier L.O.12, an all-metal type with twin engines.

The following table shows the rela-

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two of the members were chosen from among the three states through which an airway line runs. Of the others, Senators Hatch and Hiram Boren (Ore.) jointly expressed an open hostility to the way in which air transport was being supported through the postal service. Senator Hiram, in fact, was the victim of the minority coalition. In last February's Senate fight over the acceptance of the House recommendations for HAWAIIAN for the air mail, a fight ultimately won by the opponents by a vote of 36 to 30, Hiram and White were among the supporters while Hatch and Kent opposed it. 30 Senators it now to the Senate with the present session of Congress. The House committee approved in turn a series of proposals which would have provided aviation itself into two parties approximately equal.

The new investigation, which promises to be of unceasing thoroughness, will probably start early this summer and continue for several months. In the meantime, we have been in the report of the House Committee and the separate report of Dr. John D. Crane, of Harvard University, retained by the committee as a special investigator. Two documents which, jointly led to the drafting of the new Kelly bill.

The committee's first move was to go down into recent financial history and operating statistics. What they found in their inquiry seemed them to make such expenditures as first "a history of the administrative policy of the Post Office Department has been the pervasive ineptitude of the payments disbursed under different air mail contracts."

The large number of variables (in the formula for payment) introduces an element of flexibility which enables the Government to vary its payments in the different lines without theoretically disrupting the formula. Our first finding that some work in the re-allocating powers of the Postmaster General is "highly desirable." Dr. Crane in his report went still further. Most of the country's airlines and most of their needs by the committee as well, addressed themselves specifically to the policies of ex-Postmaster General Clegg, and especially to the freedom with which he had expanded and unbalanced routes and increased the expenditures paid for operating lines.

While the general use of the formula included in the report was what might have been expected the details and details had some rather less than average. The Mail report said it was possible for the first time to secure a real picture of the total usage of air mail activity and of its own divisions.

A self-supporting service

From the new calculation it developed that two transport systems handle almost exactly four-fifths of the total domestic mail traffic, and receive barely

half of the total pay. Four systems carry on the aggregate just over 10 per cent of the total traffic. The gross income to the government for the carriage of air mail in 1932 ranged from \$108 cents per lb mile (the Transcontinental & Western Day) to \$120 cents per lb mile on one of the smallest systems. Though T. & W. A. showed the lowest cost for a complete system, the lowest for a single contract route (\$126 cents per lb-mile) went to the credit of United between New York and Chicago. The committee presented in all its subsequent calculations upon the basis that the revenue to the Post Office from annual postage, above the cost of ground handling at the end, might reasonably

amounting by necessary to be regarded, or even *up to* as far as that is legally possible, *rather than* as a gross profit for the service of the air mail.

Chiefly, however, no such change in policy to give to take place without violent action from the committee that would render most difficult.

The Kelly Bill of 1933

When the committee turned from investigation to construction, they prepared the bill which was introduced under the name of Congressman Kelly. The Kelly Bill, perhaps with some minor amendments, seems to be in accord with the administration's present

policy. It makes all government operations as far as possible, and as quickly as possible, as profitable as possible. It is made possible to accept as an investment in the transportation service, were it not for the pending Senate investigation. With that hanging over the head of the air transport

business, it seems most probable that the development of air transport policy and the passage of new legislation will be based on the investigation of the House Committee reports on findings. To the committee, Congress is engaged in replacing the President with the authority to expand transportation facilities for the American Postmaster General. Here the House Appropriations Committee, in connection with the possibility of corporation, that "has well been estimated off about \$5,000,000 more, at times."

Those who drafted the Kelly Bill established themselves Kelly upon a flat determination that the air mail ought to carry its own weight economically within the next few years. With that in view, the bill provides that to rapidly as possible the present route subsidies should be converted into route warrants, under which all mileage of air mail would be paid for at a normal base rate of 0.2 cents per lb-mile. It was further provided, for exactly what reason has never been ascertained, but presumably because the committee had a very strong prejudice against large airplanes, that the base rate should not carry rates over 50 cents a mile. If those rates were to be put into effect immediately in country that could have, they would result in a decrease of at least 25



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per cent in the compensation of every existing mail contractor, and a reduction of more than 75 per cent in every mail contract line.

But the bill includes a stock-allocation. To provide a period during which passenger business can be developed to the point of enabling the lines to survive to large extent in their own right, it is provided that the Postmaster General may at his discretion pay to any operator an additional sum not to exceed 20 cents per airplane mile. The additional payment would be a subsidy on a sliding scale, to be based upon the ratio of the contractor's revenues, the money presently being derived on such a scale to would be based on any rate and would be provided all the carriers from total fixed cost subsidies. In any event, the bill provides that all such additional payments would have to terminate on June 30, 1939. Roughly speaking, to keep themselves in as good a condition as they are in now, the operators would have to increase their mail and passenger traffic to about five times the present interstate during the next five years. If that point could be reached, the bill would be for from now, but by the same ratio it would be for from now. In the case of two or three lines now handling the demand traffic, a 50 per cent increase in total traffic would be enough to permit carrying on under the subsidized regime as well as at present, even without any further increase in operation.

Even with the 20-cent rate a payment for each mile flown, it is not probable that most of the smaller operators would be a good deal worse off than they are here in the past two years. Even if they were to be paid for each mile flown, it is not probable that the payment allowed by the bill, three of the existing routes would suffer a loss of more than one-third of their total income. In comparison with the existing situation, it could be said that which they received in 1932. Such subsidies would result in some loss of income to the government, and Congress will have shown themselves friendly to an increase of the maximum subsidy allowed to 40 cents a mile during the next period. Congress is not likely to help to 40 cents the bill, but as yet passenger business is a major part in promoting growth rapidly. A study of the Government's revenue on the air mail shows the first three months of 1933 showed the rates of passenger revenues to total revenues to range from 50 to 45, with Pennsylvania Air Line having the highest percentage of passenger income. The percentage of passenger revenues available for mail during the same period ranged from 13 to 38, with T. & W. A. in first place and United Air Lines a close second.

What the operators think

The air mail operators, so far as they have expressed themselves, are in full accord with the idea of gradual elimi-

nation of subsidy and the attainment of full self-support. There is even some suggestion that they favor a total route transformation to a flat payment of 0.2 cents per lb-mile, with an even a temporary additional allowance. The operators' attitude is in line with the Washington line in April, however, some of the general plan of gradual suppression of route payments, but some suggested to any idea that would make compensation depend upon an audit of operating costs. The operators' attitude would be a subsidy on a sliding scale, to be based upon the ratio of the contractor's revenues, the money presently being derived on such a scale to would be based on any rate and would be provided all the carriers from total fixed cost subsidies. In any event, the bill provides that all such additional payments would have to terminate on June 30, 1939. Roughly speaking, to keep themselves in as good a condition as they are in now, the operators would have to increase their mail and passenger traffic to about five times the present interstate during the next five years. If that point could be reached, the bill would be for from now, but by the same ratio it would be for from now. In the case of two or three lines now handling the demand traffic, a 50 per cent increase in total traffic would be enough to permit carrying on under the subsidized regime as well as at present, even without any further increase in operation.

To bring the air-mail revenue up, and to replace the present loss when postage rates were raised last summer, the new scheme includes a contract whereby the rate on letters and packages would be based on a fixed 5-cent rate for each half-mile. The bill further calls for the creation of an air mail postal card, which would weigh about one-eighth of an ounce, and could be used for collecting both postage and the cost itself. There is a strong sentiment among the members of Congress, and among the public, for the further inclusion of a special letter-rate of the type provided in American mail lines by the Government of Great Britain, or "airletter" as Colonel Paul Henderson has named it, to be sold and carried at a special rate, and a postal card, which would be a large revenue per paid to the government than ordinary mail rates at the 5-cent rate.

Possible bill for independent
Another Kelly bill provision would extend a helping hand to a few of the smallest operators by providing for mail contracts by passenger. The Postmaster General to put air mail onto their ships, provided only that their routes do not cross the lines, which some operators may have been given. The cost on such cases would be a flat 0.2-cent-per-lb-mile, with no possibility of extra allowance. Another suggestion prohibits any

company having on its own route without from holding any interest in the contract line of any other company. Clearly the attempt is that each airline, even though it is as far as United Air Lines, T. & W. A., or American Airway, but have a single contract to carry its entire service. The prohibition on cross-holding ownership is presumably aimed at future attempts to acquire ownership, would not be a first step to prevent holding companies at all. The Housekeeping Senate report, which is a substantial recommendation of the responsible committee in the House of Representatives has indicated no desire to destroy the existing business of the operators of the industry.

It is rather a remarkable fact that the provisions of the new bill, based upon a careful consideration of six years of operating experience with the contract mail, should be so very clearly stated in some respects to the Kelly Bill of 1932 and its amendments. The Housekeeping Bill of the same year. Eleven years ago Congress Kelly introduced a bill to permit of the letting of air mail contracts for a period of a maximum rate of 0.3 cent per lb-mile a figure, which, as experience has made evident, would have been far too low in period of rising air fares, and it is somewhat too low even now. The Housekeeping Bill raised the proposed rate to 0.2 cents per lb-mile, or exactly the same allowance, but, being recommended now, eleven years later. It had as a supplementary feature, however, a sort of authority for the Postmaster General to send delivery free of charge by air at a considerably lower rate per lb-mile a provision that may well be somewhat of a surprise to those who, ten years later, though it is necessary to reconsideration at the present time.

Corrections

ATTENTION! has been called to certain inaccuracies contained in model number 4 on page 108 of the May issue of AVIATION. The illustration of the lower left hand corner should be replaced 4-750-E instead of 4-500-E. The 10th and 11th columns of the table should be 10-200-E and 11-200-E of the K-101 series.

Through a difference in interpretation of the actual data of the P-100 P-100 table appears to have been incorrectly as 700 lb in the Directory issue. The corresponding table appearing in the 1933 issue of AVIATION should be 700 lb in the Directory issue.

It should be noted that the maximum speed given on the three-hour drawing of the P-100 is on page 125 of the May issue of AVIATION, which shows the lowest speed, as pointed out in the text on the same page. The current top speed is 145 m.p.h. or 125 m.p.h. depending on the Continental 2100 hp engine.

Although it is a matter of general information that controllable pitch propellers provide a definite improvement in such performance factors as take-off, climb, and single-engine cruising of multi-engine ships, it is not always realized that these are increases cruising speed. The series of tests herein described, which were conducted by the research division of the United Aircraft & Transport Corporation at Rochester Field under the author's direction, were inherently convincing in results in a decision to install controllable pitch propellers on the last 30 of the 60 Boeing Model 247 transports now under construction for United Air Lines.

Controllable pitch propellers in transport service

By Charles Hugh Chaffield

Rochester Division, United Aircraft & Transport Corporation (of Connecticut)

ALTHOUGH the speeds of transport airplanes have been increasing for many years, it is only relatively recently that these gains have been attained more by reason of airplane drag than by increase in engine power. So long as power limitations were being progressively reduced, all the airplane performance characteristics except range and endurance were automatically improved, but now that higher speeds are accompanied by stationary or even increasing power loadings, the limitations of the fixed-pitch or ground-adjustable propeller have become more and more apparent.

The wide disparity between take-off and maximum speeds at a medium altitude means that a propeller designed for the latter condition is not particularly well suited for the former, for its high blade angle tends not only toward a reduction in efficiency at low forward speeds, but also to so high a torque coefficient that the engine is taken over at full throttle, with a substantial margin to reach its rated speed, and consequently cannot deliver its rated power. Thus a relatively low thrust per rated horsepower may be realized with a rather high airplane weight per horsepower, with the result that satisfactory in take-off losses economically to be derived. If a speeded-up engine is used which requires a throttle-stop at low altitudes, the take-off performance is further impaired.

To a lesser extent this is true of the efficiency of the power plant in advance of climb because the characteristics that at the maximum speed propeller deliver intensify to such a degree that the desired forward speed of the climb condition

in a multi-engine airplane, the condition of flight with one engine out of action is unobtainable in that of climb, and the reduction in winged extent by stopping an engine results in a loss of thrust available from the remaining engines.

But the controllable pitch propeller affords marked improvement in take-off and climb, and in flight with one dead engine is practically neutral. It has perhaps not been so widely realized, however, that the ability to change blade angles in flight often makes possible also an improvement in cruising cruising speed.

Now that a controllable propeller of the controllable type is available, it is felt that there may be some interest in order data on the performance of a modern high-speed airplane thus equipped. The material in question is a Boeing Model 247 transport airplane with two three-blade Wasp engine S1D1 engines developing 580 hp at 2,200 r.p.m. at 5,000 ft.

To investigate the effect of controllable propellers on the performance of an engine airplane, flight tests were made at Rochester Field in Hartford in April and May of this year. Seven previous tests made by Boeing Airplane Company at Seattle with several designs of Hamilton Standard fixed-pitch propellers had shown that best all-around results were obtained with three-blade propeller 9 ft. 1 in. in diameter, these were accordingly used in the Hartford tests to establish a base for judging the gain in performance due to the use of the controllable propellers. These test propellers were of the new three-blade hydro-aerodynamic type 9 ft. 2 in. in diameter, with five blades.

The weights of the two-blade controllable propellers was very nearly the same as that of the three-blade fixed-pitch propellers.

It will be noted that while the tests of cruising speed and of climb with two engines were made at the unusual gross weight of 12,600 lb., those of take-off were conducted at the low gross weight of 10,100 lb. For the tests of single-engine climb the airplane gross weight was less with the fixed-pitch than with the controllable propeller, but for each of the other comparative tests of controllable and fixed-pitch performance, the gross weight was kept constant. The test results follow.

Table 1

With an airplane gross weight of 11,100 lb. and an engine installed power of 34 ft. 11 in. the comparative figures are shown in Table 1.

Propeller	Controllable three-blade initial	Fixed-pitch three-blade initial
Take-off distance (ft.)	180	200
Take-off time (sec.)	11.2	11.8
Take-off rate (ft./sec.)	15.5	15.3
Best climb rate (ft./min.)	1,040	1,030
Rate of improvement with controllable propeller	20 per cent	None

The take-offs were all made from the gross weights of Rochester Field, and the results have been reduced to the sea-level condition, although at sea level during the take-off tests was the wind velocity over 7 m.p.h. The figures given are in each case the average of at least three take-offs. The broken were set at the beginning of each run, as this method of testing has been found to give



Hamilton Standard controllable pitch propeller installed on the Boeing 247

more consistent results than that of releasing the brakes before the throttle is opened.

Climb with both engines

The gross weight of the airplane was 12,600 lb. for the two engine climb, and the engines were operated at full throttle at the initial altitude of 5,000 ft. Table 2 gives the climb data.

Propeller	Controllable three-blade initial	Fixed-pitch three-blade initial
Rate of climb at 5,000 ft. ft./min.	1,040	1,030
Climb in three (ft.)	3,000	2,900
Best climb rate (ft./min.)	11.5	11.3
Engine speed in climb at 5,000 ft. r.p.m.	2,150	2,160

At 5,000 ft. the engine speed was 2,150 r.p.m. at 11 m.p.h. at 10 m.p.h. at 11 m.p.h.

The propellers of both climb were determined by preliminary smooth climb in the wind tunnel, and the figures given above have been reduced by Duff's method of S.A.C.A. standard atmosphere.

Climb with one engine

The one-engine climb tests were made with which up; one engine was throttled back, and the other was run at full throttle. Tests were first made to determine minimum rates of descent at two altitudes above the one-engine climb, and the rate-of-descent curves plotted from these data were produced to give rates of descent. The one-engine climb that obtained was verified by level flights at that altitude. The figures are shown in Table 3.

These results have been reduced to give

NACA standard atmosphere by Duff's method. The blade angle of the controllable propeller was 13.9 deg. at the 42-in. radius, and the speed of the engine at which the throttle was open was 2,000 r.p.m.

Cruising speed

The use of controllable propellers permitted a high-speed climb using 20 in. level flight at 5,000 ft. in the engine inlet static pressure was 25 in. Hg. absolute at an engine speed of 2,000 r.p.m., and the engine power was increased 75 per cent of the rated power.

Propeller	Controllable three-blade initial	Fixed-pitch three-blade initial
Engine speed (r.p.m.)	2,000	2,000
Engine power (hp)	1,040	1,030

At 5,000 ft. the engine speed was 2,000 r.p.m. at 11 m.p.h. at 10 m.p.h. at 11 m.p.h.

The propellers of both climb were determined by preliminary smooth climb in the wind tunnel, and the figures given above have been reduced by Duff's method of S.A.C.A. standard atmosphere.

Climb with one engine

The one-engine climb tests were made with which up; one engine was throttled back, and the other was run at full throttle. Tests were first made to determine minimum rates of descent at two altitudes above the one-engine climb, and the rate-of-descent curves plotted from these data were produced to give rates of descent. The one-engine climb that obtained was verified by level flights at that altitude. The figures are shown in Table 3.

These results have been reduced to give

practicable for actual cruising operation less than that as it is undesirable to exceed either of these limits, the raising power with fixed-pitch propellers is governed by the first limit reached, that is, by the lowest engine speed.

The maximum speed at 5,000 ft. with both throttles was only 1 m.p.h. less with the controllable propellers than with the fixed-pitch propellers; this is considered important in view of the gain in cruising speed. It would, of course, be possible, if the gain in cruising speed were not desired, to set the high-light portion of the controllable blades so that the engine would run at their rated speeds at full throttle, thus obtaining the maximum speed.

Another question is that of obtaining the higher value of the high-speed climb, but raising the engine speed above that, thus obtaining the engine speed actually given by the fixed-pitch propellers, but with a definite improvement in fuel economy.

As a result of the improvement in airplane performance shown by these tests, the controllable propellers are to be installed on the last 30 of the 60 Model 247 transports being built by Boeing Airplane Company.

Table 4

Propeller	Controllable three-blade initial	Fixed-pitch three-blade initial
Engine speed (r.p.m.)	2,000	2,000
Engine power (hp)	1,040	1,030
Engine speed (r.p.m.)	2,000	2,000
Engine power (hp)	1,040	1,030

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The propellers of both climb were determined by preliminary smooth climb in the wind tunnel, and the figures given above have been reduced by Duff's method of S.A.C.A. standard atmosphere.



Above: A new setting for the annual photograph—the Oldfield wheel house.

Left: One of the airplanes used to test a device for use in testing in flight.

Below: A special installation on the tail of an airplane designed for the flight test section.



At Langley Field

Three views of the Eighth Annual
Engineering Research Conference

The N.A.C.A. receives callers

Standard procedure is modified at this year's conference to accommodate additional guests and present more subject matter

THE ANNUAL engineering conference of the National Advisory Committee for Aeronautics has expanded its class-room around a flight test making it possible to check not only the results but the accuracy of the testing equipment and methods.

The morning session

The morning session began at 10 a.m. with a summary of the work of the Aerodynamic Division by its chief, Mr. Elmer W. Miller. Mr. Miller called attention to the recent increase in aerodynamic efficiency due to drag reduction and mentioned that, whereas airplanes previously wasted one-third of their power in the production of turbulence, now only one-fourth of the power available is lost by imperfect streamlining. Outlining the methods of analyzing drag, he discussed first the problem of wing-fairing interference. A study of the position of the wing with respect to the fuselage showed the drag to be a maximum when the wing was bent just above the mid-wing position. An interference value of —4 per cent was found in comparison with a value of —4.50 per cent with the wing at the bottom of the fuselage. With the best fit, one having a large radius of curvature, a value of —15 per cent was obtained. Any effect from wing was found to reduce interference to zero. A study of airfoil nose curvature showed that small changes in radius made but small differences in lift. A sharp-nosed airfoil, having a basic radius of 1 of the chord, showed a reduction in maximum lift of about 30 per cent.

A new propeller section, designated NACA 206, with sharper nose and a maximum ordinate further back than that of the Clark Y (40 per cent from leading edge), has been tested in the high-speed tunnel. When the ratio of the airspeed to the speed of sound was 3.5, this section had one-third less drag than the Y test over the drag element at a value of the same ratio of 0.7. A systematic investigation of the form of NACA airfoils is under way as a joint project of the Aerodynamics and Power Plants divisions, and the work on wing-fairing has been continued in the Propeller Research Tunnel. In general, better installations have been found to be better than the previous ones, showing freedom from the standard of efficiency and drag is that with the

propeller directly ahead of the leading edge. In this position a combination of cooling and turning reduces the power required to overcome the drag to about 1 per cent of the output of the engine. The drag of stalling, stopped, and free-wheeling propellers has been studied and the advantage of the aerodynamic pitch feathering type is clearly shown.

The cyclopropane

"Lift without speed" was the test of the opening portion of the talk by J. W. Crowley, Jr., chief of the Flight Research section, who has been conducting a special study of rotating wing systems of all types. It was Mr. Crowley who introduced to the audience the "cyclopropane," the paddle wheel rotor design, patented in this country by Elmerand W. Starr and in Germany by Dr. Adolph Reibhahn. Computations indicate that a 2,000 h. p. engine utilizing this principle, with constant velocity rotor and main, would attain a speed of approximately 100 m.p.h. with an intake of 300 lb., assuming that 60 m.p.h. remained in the propeller losses. The power required would be a minimum at 45 m.p.h. and increase 60 per cent at 50 m.p.h. and 90 per cent at 100 m.p.h. The power required at 100 m.p.h. would be 780 h. p. per minute and the maximum rate of climb 1,500 ft. per minute.

To compare the effects of stillability on pressure recovery, as a part of the opening investigation, three vertical and three horizontal pressure were tested and it was found that the pressure in the vertical was increased 30 per cent by tilting the horizontal tail to the top of the rotor and 21 per cent by moving the rotor further back a distance of one-tenth of the chord.

Two tests and its effect on lift was also studied and it was found that by moving the tail of the air and carrying the air forward improved the wing's quality. The change in distribution and position of the air from the undisturbed to the highest values reduced the number of air molecules per recovery from an indefinite number to 11.

The effect of control surfaces on probable load factors has been investigated and a list has been given that the most favorable results have been obtained. When the lift force was required to obtain 60 and 100 ft. the max-

mean normal acceleration in recovery from a dive was 4.66 G. When the time was 10.16 seconds in recovery was 7.72 G. From 50 ft to 325 ft, one-half heaviest engine table difference. For load factor in inverted maneuvers the lowest factor was found to be the physiological resistance of the pilot—(4) being the maximum to which a pilot will subject himself.

The results of the simulation of the V-2 corridor on a number of commercial transports have given some clue as to the frequency of gusts. The maximum normal component of a gust recorded (in 1,500 hours flying time) was less than 20 ft a second. A gust producing a vertical velocity of 25 ft per second would produce a load of 3.5G when cruising at 120 mph and 5G at 200 mph.

Power plant program

Reviewing the accomplishments of the Power Plant division Mr. Carlton Kemper told part of the problem of engine of double-row jet air-cooled engines. Studies have been made of the fan packs, which and thickness, the pressure difference and quantity required of cooling air, and the problem of cooling the rear cylinders in two-row engines.

A study of the effect of the distance from fan to bulk and temperature distribution showed that the closer the bulk to the cylinder the more effective the cooling. Better must be at least within 2 in. of cylinder.

Progress has been made in the adaptation of safety flap (EAS flap, F flap) and it has been possible to simulate the same engine and configuration that obtained for engine problem with normal water arrangement. With EAS flap developed and EAS in memory having it has been possible to obtain a 130-MPH of 200 ft per second and a consumption of 40 lb per hp-hr starting from cold by the operation of position into the manifold.

The single-cylinder, two-stroke-cycle compression ignition engine modified in last year's conference has been tested and put in service. The cylinder has four ribs on the top and six at the bottom as the bottom is a swirl, so that the long lost at the cylinder has values up to 115 lbs have been obtained with fuel consumption of 87 lb per hp-hr. At a 55 K.P. of 180 lb per hour, the fuel consumption was 45 lb, per hp-hr.

Mr. Kemper concluded by outlining work in which hydrogen was used as fuel for compression-ignition engines with the ultimate objective of increasing payload and eliminating the water injection system weight to lighten. The functional change in the type of fuel is necessary for the use of hydrogen and it is estimated that in an aircraft of 6,000-lb in lb employing zero hydrogen could be carried by helium

only, but having hydrogen in the engine, an increase of 25 per cent in weight and 42 per cent in payload could be obtained.

The icing bench

Within the fortnight preceding the completion the opening of the new icing bench at Langley Field was increased from 15 to 50 mm per day by the installation of wire screens, containing essentially of flat rectangular screens set just below the water level and at intervals close the sides of ducted. These reduce the time necessary for the installation to simulate after each run. Mr. Tom Truitt, who is in charge of the bench, outlined the program, changing from the effect of rough and loose on the resistance of a hull model. The heavy resistance falls off with an increase of hull diameter, while the light-weight resistance varies in the opposite way with particular respect to the changes in beam. A hull form designated NACA 4 was found to be 10 per cent better than the 1000 (March) model. It has the same section as the March but less area at the bow in profile and greater area.

A visit to the laboratory

An excellent opportunity to visualize the functioning of the cyclotron was afforded at the main flow tunnel at the variable density building through the efforts of E. M. Jacobs and John Stuck.

In the power plant section the new two-stroke cycle engine-cylinder design which has run 150 hours at speeds up to 2000 rpm, and the apparatus for the measurement of hydrogen rate a some possible liquid engine with direct.

The study of airplane fuel demand was discussed by E. V. Skala in the flight test section. Other exhibits in the flight test section included the preliminary results of a systematic study of blood areas in cockpit designs. A reference book on the pilot's eye at the center, and having parallel of horizon and meridian of longitude on its surface, in the form of the system together with a sighting device which is placed at the eye position. Results so far indicate that typical tractor engines have been found to 75 per cent.

An interesting feature of the charts were arranged in a display row in the Progress Research Tunnel and most of the data was taken by Donald Wood, in the work on models and principles locations. The drag at a stream-line wheel was found to be 25 per cent higher than that of an airfoil. The drag while the drag of a low-pressure tire was 2 per cent higher, and at an extra-low-pressure tire 20 per cent higher than that of an airfoil.

In a representative landing gear study in the Pull-Down Tunnel under the direction of South Dr. France it was determined that the resistance of the Lado-

but Altair landing gear in extended position is twice that in its retracted position. The system of landing gear with made no difference in drag. Resistance of rivets on aerial surfaces also have been studied.

After luncheon the audience participated in hear Fred E. Weid.

Increased lift, lowered control

Maximum lift coefficient as high as 3.17 have been obtained for the Fowler wing with flap retracted, but the addition of a Handley-Page slot in that arrangement resulted in less moment in cranes. By developing the slot through modification of the slot, a value of 3.52 was obtained and the additional effect of other slot at the rear resulted in an increase to 3.75. Through the use of the Wright auxiliary flap at 45 deg the lift coefficient was raised from 1.17 to 1.81 (based on total area). The low auxiliary airfoil forward had a chord of 1 in to 13 of the wing chord and was level with the upper surface of the wing and located about one-third of the chord length ahead of the leading edge.

Vicinity of the chord of a split flap resulted in a mild increase in maximum lift up to 20 per cent of the wing chord without any appreciable gain thereafter. Cutting away the flap from the tip resulted in lift reduction gradual at first, increasing rapidly beyond the semi-span point. Cutting off from the coast is rapid 18 reduction at first, becoming more gradual later.

Increasing moments observed in flight corresponded to those measured, around the wing root, rather than body axis. No ordinary moment was satisfactory in one at high angles of attack or at about the end and some were controlled above the stall. Spoilers were found satisfactory in this region, but there is an approach by its other action. With a conventional spoiler, the rolling moment was higher at the stall than throughout the range of angles and the yaw was high end of engine nose.

Reference to the tip of spoilers Mr. Wood described flight tests in which the ordinary airfoil over a maximum angle of attack and the maximum moment as an increase, immediately. This was compared with the wing type which gave an angular velocity of 100 rpm in 15 seconds, and the same for the first 5 seconds, and the remarkable spoiler which showed values of 11 rpm in five seconds but required 7 seconds to show any angular velocity.

With upper surface spoilers and split flap the roll was normal and the yaw much improved and with external spoilers attached to the upper surface to about 25 per cent of the chord, good roll and positive yaw were obtained in 5 seconds.

Following the talk attached to the trail by the edge of the wings was tried and found to be very effective up to 15 deg, but the control was lost above that point.



View of shop from end of assembly line.

Engine overhaul divorced from engine manufacture

A survey of United Airports of Connecticut, Inc., engine servicing shops at Hartford

ACTING on the theory that the owner of engine equipment is responsible for its upkeep, practically every company in this country and abroad has installed his own shops for servicing and overhaul of airplanes and engines.

There are few exceptions. In France, for example, Air Union, an operating company, is so closely associated with the manufacturers of its equipment (the Farman Works) that except for routine servicing and overhaul, all engines and engines are sent at stated intervals to the shops in which they originated the complete inspection and overhaul. Elsewhere in Europe and in the United States, however, the trend has been to the opposite direction. There are a dozen or more well-equipped shops scattered about the country (particularly at air centers by transport lines) where it is possible to overhaul and to completely rebuild, if necessary, all types of radial engines. The engine manufacturer usually takes it as his duty to advise the customer where he can

with the most of their equipment have obtained extensive servicing operations to keep in touch with engine units as long their operators go to factory numbers at all times. Beyond such co-ordination of services, however, the manufacturers of engines and engines have found it desirable to divorce the purely maintenance and the purely servicing functions within their own factories at its equipment (the Farman Works) that except for routine servicing and overhaul, all engines and engines are sent at stated intervals to the shops in which they originated the complete inspection and overhaul. Elsewhere in Europe and in the United States, however, the trend has been to the opposite direction. There are a dozen or more well-equipped shops scattered about the country (particularly at air centers by transport lines) where it is possible to overhaul and to completely rebuild, if necessary, all types of radial engines. The engine manufacturer usually takes it as his duty to advise the customer where he can

As a vivid example of a step of this sort is to be found in United Airports of Connecticut, a subsidiary of the United Aircraft and Transport Corp., associated with the Pratt & Whitney Aircraft Company at Hartford, Conn. This company not only inspects engine support operations at its main office, but more particularly functions as an overhaul unit for Pratt & Whitney, leaving that organization free to devote its entire time to the problems of manufacturing United Aircraft's main shops are located on Remondier Field, adjacent to but not connected physically with the engine manufacturing company. At the head of the inspection there is R. L. Whelan, veteran pilot and engine maintenance man of long experience. The unit at East Hartford maintains its own largest shops and personnel, and only in very exceptional circumstances is it necessary to call upon the manufacturing division for any sort of work on engines.

The United Airports building on Remondier Field is a modern brick and steel structure, whose main portion is the large garage—100x125 ft—with outdoor bays to handle the largest ships now in service. The two ends are filled with electrically operated cranes

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THE BUYERS' LOG BOOK



Kollsman fuel gauge

KOLLSMAN Instrument Company, Inc., makers of sensitive altimeters, now widely used in transport and service airplanes, have lately developed a fuel gauging system which eliminates many of the difficulties encountered in the installation and use of the ordinary hydromatic type. In place of the usual hydromatic cell installed in the fuel tank from the top, the Kollsman system substitutes an outside tank cell and safety valve and an automatic protecting valve at the top of the tank to prevent fuel entering the lines to the gauge on the instrument board. This is important, because gauges are sometimes damaged by fuel entering the lines, or by excessive tank pressure due to filling from high pressure gauges. The outside tank cell is entirely external to the tank itself. It is unnecessary to engineer any great care or accuracy in installing it (as any small leak in the tank, for example) and it is conveniently adjustable regardless of tank size, shape or volume. In the event that the connecting line is broken or disconnected for any reason, no gasoline can escape, since a valve automatically closes until the fuel system is again airtight. A special vent maintains the pressure in the valve continuous line and so up to the fuel in the tank at tank value. In the Kollsman system a fuel quantity gauge and pump are installed in the instrument board in the usual manner. The connections for the gauge and pump are shown in the accompanying sketch.—*Aviation*, June, 1933.

Aircraft towing signs

WITH the increasing popularity of aerial advertising by means of large signs towed behind aircraft (see

Aviation, December, 1932, page 493), Air Transport Company, Inc., Garden City, L. I., N. Y., has developed and placed on the market a complete line of signs for towing purposes. They may be towed behind subsonic, stall aeroplanes, or any average airplane. The signs are of a new type which permit the quick changing of letters.—*Aviation*, June, 1933.

Fuel safety valves

A SERVICE which automatically shuts off all the flow of gasoline to the chamber when the ignition switch is turned off has been developed for aircraft use by Westinghouse Electric & Manufacturing Company of East Pittsburgh, Pa. The primary purpose is to eliminate fires from broken fuel lines. The valve which was developed by Mr. L. G. Riley, a Westinghouse engineer, consists of an inner steel core and an outer bellows tube which forms an integral part of the machine line. When the ignition switch of the engine is turned to the "off" position, current flows through a coil of wire wound around the outer tubing and exerts a magnetic pull upon the inner core, pulling it open against spring pressure. When the system is cut the flow of current ceases and the spring forces the core close, thus sealing off the fuel line. A new type of flanged, flared end joint eliminates leakage. The valve has no stuffing boxes to pack and

no glands to remove, and is said to remain leak-proof throughout its useful life. Valves can be arranged to operate on any voltage from 12 to 350 and are available in three sizes to handle liquids or gases in pipe lines ranging from 1/4 to 1 1/2 in. diameter. The smallest now weighs 3 1/2 lb. complete and the largest 16 lb.—*Aviation*, June, 1933.

Fairchild five-lens camera

COINCIDENT with a report that the Bureau of Geographical Exploration of Harvard University had completed a photographic record of the entire State of Massachusetts since the



Fairchild five-lens camera

achievement of a special five-lens aerial camera by Fairchild Aerial Camera Corporation used for the survey. The Model 13A aerial camera is receiving five cameras installed into one. The central camera takes a vertical photograph, the other four cameras take oblique angles. Each camera is equipped with a film magazine with a capacity of about 200 exposures and the film is all five cameras is advanced simultaneously by a turn of a crank. The shutters of all five lenses are also tripped simultaneously by a single lever. After development the exposures from the central camera are contact printed, and those from the oblique cameras are reloaded and brought to the same scale by means of a Model B-7 transformer printer which is a part of the camera ground equipment. The five prints are then mounted accurately and assembled in the form of a Maltese cross. The new camera weighs many times more than in a single exposure that the Model 12B type, and therefore naturally reduces the time required to survey a given area. The width of strip covered by the TSA camera is about 1 mile by every 1000 ft. of flying altitude. From 14,000 ft. one photograph covers 136 square miles.—*Aviation*, June, 1933.



Westinghouse fuel valve

OUR ANSWER IS: WE NOW DO!

PILOTS wanted it. They said "make it easy for us to carry an extra can of Kendall with us so we can always be sure of having it when we need it. . . no matter where we are."

And so the aviation grades of Kendall went into refinery-sealed cans . . . and we hope you always carry a few extra with you. They're easily opened by pushing a couple of holes in the top with a screw-driver. And you can pour directly from the cans into your oil tank without loss or bother. *That's the way!*

We could add, of course, that these refinery-sealed cans also give perfect protection from dirt and dirt that can so easily clog every airport. The oil goes into your motor clean and pure, just as it came from the refinery. Most airports now can serve you with Kendall in refinery-sealed cans.

KENDALL
THE 30 HOUR OIL

MERELY MAINTAIN THE PROPER OIL LEVEL

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EFFECTIVE Rocker-arm Lubrication is an Accomplished Fact

TEXACO MARFAK GREASE has supplied a pressing need in aviation. Endured rocker-arm assemblies present a difficult lubrication problem. Temperatures and bearing pressures are high. Ordinary greases rapidly lose their protective qualities. The safe life of the lubricant is short. Texaco Marfak Grease is unique.

It is a pure mineral oil lubricant without a trace of artificial filler of any kind. It is extremely resistant to heat and pressure, clings to the bearing surfaces and stays soft. It will not throw off. Texaco Marfak Grease lasts many times longer than other known lubricants for this purpose.

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Marfak Grease 100 per cent. It is used and recommended by the leading air transport operators, by engine builders and pilots from coast to coast. Texaco Marfak Grease and other Texaco Aviation products are available at principal airports and landing fields everywhere. Write The Texas Company for full details.

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THE Truth about Airplanes

ALL FACTS... NO FICTION



Anybody thinking about buying an airplane ought to know all angles of the competitive situation. So here are a few facts about price—about performance—about service—about performance based on sales and engine-test figures.

Fact No. 1. Official government engine-test figures prove there were more four-place Waco's sold in 1932 than any other make. Counting both open and closed planes, Waco outsold its nearest competitor better than 2 to 1. That's fact—most facts. And you'll have to admit it takes a good ship to win such overwhelming popularity.

Now for Fact No. 2. Waco's price has built a name will build a ship to meet

competitive prices. Why make the world's finest ship a second-rate job in order to sell it for a little less money? If we did that we wouldn't continue long as first plane.

And here's Fact No. 3. Performance is one of the principal features in an airplane. And you have to pay for performance in anything you buy. Waco gives you 30 miles per hour better top speed and 25 miles per hour better cruising speed. That means a lot of money in a year's time. Then there's the safety factor. Waco's landing speed is 14 miles per hour less. There are hours when this feature alone is worth many times a Waco's original cost. In other words, the performance

you pay for in a Waco is not obtainable elsewhere. That's a Waco in a "good buy" at the price asked for it.

This means you can't buy a better airplane than a Waco. Naturally it costs more to build this kind of a ship and so have to charge more for it. You can get a plane for less money than what a Waco costs. But you can't get Waco performance—and safety—for less than Waco's price.

A word on plane distribution: Waco airplanes are sold by America's only actual independent dealer organization. There may be some for you in the South. Write for particulars. The Waco Aircraft Company, Troy, Ohio.

HIGH SPEEDS...YIP FOUR-PLACE CABIN WACO

These stream-line design 4-Place Cabin Waco's offer 40 miles an hour cruising speed at economical cruising altitudes of 10,000 to 12,000 feet. In less than 300 yards they can take you any other place. It costs less to operate (payroll) and can be repaired at half the cost (depend on parts cost for moment) in all parts, an feature.

WACO AIRCRAFT IN AIRCRAFT REGISTRATIONS





The new Pioneer Electric Tachometer, Type 639, thoroughly reflects true Pioneer ideals...to evolve new ideas and through concentrated effort to produce honor-built instruments in advance of demand.

After many years of development, in conjunction with the U. S. Navy, this new instrument is presented with several important advantages in addition to those commonly found in this type of engine-speed indicator. Comprising two units...a two-phase alternating current generator and an indicator of the induction disc type, this principle is entirely new in application and definitely eliminates the difficulties experienced with former instruments of this type. There are no brushes or moving contacts of any kind. No shock absorber or special mounting is required for the indicator. The hand, which is remarkably free from vibration, operates over a 350 degree of scale arc, thus providing an open scale not found in any other electric tachometer. Despite extreme conditions, accuracy is assured by Pioneer's method of compensation for temperature variations. Reliability is materially increased and maintenance reduced by the provision for radio shielding and operating the instrument without commutator, slip rings, or moving coils. The new Pioneer Tachometer, Type 639, is offered with the reputation and integrity of Pioneer to back it.

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Announcing the New Speedy STEARMAN

Model 80

W.C.M. is a new type biplane of the type known as the 80. It has a top speed of 125 m.p.h., a cruising speed of 85 m.p.h., a range of 4,000 miles, a service ceiling of 10,000 feet. It is built by the W.C.M. Aircraft Co., Inc., of New York City. It is a single-engine biplane with a fixed landing gear. It is a very fast and reliable aircraft. It is a very good example of the new type of biplane. It is a very good example of the new type of biplane.



AGAIN Stearman steps out. This time

it's the Model 80. You will want to know

more about this new, fast model. Strictly.

—it's a Stearman. You know that means experience, engineering, accurate

workmanship, vigorous performance. The Model 80 will

soon be at home on the skyway of the country... because Stearman, backed by United, reaches nation-wide.

The Stearman, Model 80, is being used in sales promotion work by Mr. John L. Wells, Jr., of Oak Park, Illinois.



Model 80 — 125 m.p.h.
Cruising speed 85 m.p.h.
Range 4,000 miles
Service ceiling 10,000 ft.
Climbing 100 ft. p.m.

Hamilton Standard Controllables For Boeing Transports



After exhaustive tests, Hamilton Standard Controllable Propellers have been ordered for installation on the new low-wing Boeing Transports for United Air Lines.

The use of controllable propellers demonstrated substantial improvement in take-off, climb, cruising, and single-engine performance.

HAMILTON STANDARD CONTROLLABLES
ARE IN USE BY, OR ON ORDER FOR,
THE FOLLOWING AIR LINES:

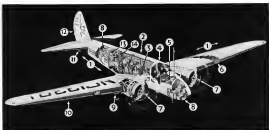
*Pan-American Airways
Pan-American-Grace Airways
Transcontinental & Western Air, Inc.
United Air Lines*

*Deutscher National Aviation Corp.
Deutsche Luft Marine, A. G.
Royal Dutch Air Lines (K. L. M.)
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HAMILTON STANDARD PROPELLER COMPANY - EAST HARTFORD - CONNECTICUT

BOEING 247!



Responsible for the amazing speed, strength, payload, comfort and operating economy of the new Boeing transport are the following features:

- 1 All-metal streamlined construction to moderate speed and approximate a minimum of resistance and weight. All main parts of one 247 interchangeable with those of another, including body sections, wing sections, etc.
- 2 Spacious cabin, with an interior length of 20 feet and height of 6 feet. Equipped with heating-cooling system, music and individual ventilation, dome lights, individual reading lamps and ash trays. Insulated against both noise and temperature conditions.
- 3 Wide upholstered seats with adjustable reclining backs. Spaced 40 inches apart to provide ample leg room. Each chair has a window.
- 4 Pilot's compartment with dual controls and complete assembly of flight and engine instruments. Standard eight group equipment.
- 5 Compartment for battery radio telephone equipment which makes possible ground-plane communication as well as the reception of weather reports and direction radio beam signals.
- 6 Shock absorber mounted on the main landing gear, designed to withstand a load of 36,000 lbs.
- 7 Two 225-horsepower Pratt & Whitney Wasp engines, supercharged to 3000 feet. Three-bladed Hamilton Standard propellers. Engine rim vents.
- 8 Compartment for rock braggas and engine. The main pit is 60 cubic feet in size, and the rear compartment 40 cubic feet.
- 9 Retractable landing gear, equipped with shock absorber, hydraulic brakes and large, low-pressure tires. A differential of approximately 50 mph in speed is possible with the gear retracted.
- 10 Thirty-seven inspection doors, strategically located, as well as emergency wing tips, leading edge of wing and tail cone, to facilitate inspection and maintenance.
- 11 Full-revolving tail wheel, equipped with shock absorber and dampers.
- 12 Encapsulated special leading edge rubber, elevator and aileron flaps, which provide for trimming the plane in flight and for correcting unbalanced propeller thrust.
- 13 Streamlined canopy. Seats are covered with.
- 14 Landing.

This airplane on display in the dome of the Transportation Building, Century of Progress Exposition, Chicago. Also now on order at United Air Lines. Detailed specifications on request, Boeing Airplane Company, Seattle, subsidiary of United Aircraft and Transport Corporation.



BOEING has always built
tomorrow's airplanes today!



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Above: One of the new Curtiss Condor airplanes, carrying 10 passengers, 2 pilots, a stewardess, and 600 pounds of mail, baggage and express at a top speed of 115 miles an hour over the new "Yellow Route" between Chicago and New York, via Detroit and Buffalo. Flying time, slightly over 5 hours; the planes are equipped with two 700 h. p. Wright Cyclone Engines, which use B.G. Spark Plugs exclusively. Each plane is equipped with receiver radio telephones.

Below: One of the new twin motor Sikorsky Amphibian lightships, carrying 6 passengers, 2 pilots, express and baggage. These planes are in service between Detroit and Cleveland. Flying time, 2½ hours. Each plane is equipped with two 425 h. p. Wasp Engines with B.G. Plugs in regular equipment. Each plane is also equipped with receiver radio telephones.



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